

# LPT BACKUP DESIGN

## (I) ROTOR SHOCK INSTABILITY STUDY

By Reed Chang / ITEC 1-1-85 Phoenix

張瑞釗 / 引擎合作發展國外小組

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## 1. INTRODUCTION

This report is describing the LPT rotor shock instability problems in the worst case of LPT operation.

The following is the analysis procedure:

1) LPT AERO requests performance group to offer the performance analysis cycle sheets corresponding to the worst cases of LPT operation.

2) Using TAPS program, AERO must check out the cycle which has the maximum rotor exit critical Mach no. and maximum blade axial force from the AERO'S and MCD' s points of view respectively.

3) According to the cycle sheet selected and the blade tape files from AERO, MCD analyzes the vibration problems and determines the maximum deflection blade coordinates for aerodynamic analysis.

4) AERO group compares the deflected blade and the original blade, and calculates how much the throat area and exit critical Mach no. varied when flutter phenomenon happened.



5) It must be determined whether the rotor blade needs to be redesigned or not.

## **2. PERFORMANCE ANALYSIS**

As previously stated, this analysis is requested by AERO, and needs to be done by performance group.

The purpose of this analysis is to find the worst cases for the LPT operation at peak aerodynamic loads such as the maximum rotor exit critical Mach no., maximum flow rate, and maximum pressure ratio or the combined cases of the above parameters.

Appendix I shows some of cycle sheets offered by performance group. The reason why we only choose these cycles is that the maximum rotor exit critical mach no. will happen when LPT operates within these cases (i.e. with maximum nozzle area as shown on cycle sheets).

## **3. STREAMLINE ANALYSIS**

Basically, AERO needs to run TAPS at the cycle far from the design point, say, the higher altitude to get the worst case. But, from the MCD point of views, the cycle the cycle that has the maximum blade axial is required.

Therefore, AERO needs to choose a cycle with lower altitude because the axial force of blade increases with the inlet pressure (at station 4.5), which increases when the altitude is reduced as cycle sheets indicated. And it makes no much difference about rotor exit critical Mach no. throughout all the altitudes for

aerodynamic analysis.

Using TAPS program, AERO obtained the maximum rotor exit critical Mach no.  $M_1^*$  of 1.498 with the maximum nozzle area, and the corresponding tape files were transferred to MCD.

#### **4. EVALUATION OF LPT BLADE TO AN INSTABILITY PROBLEMS (MCD)**

MCD must evaluate the LPT blade design to a shock instability problem. What AERO group needs is the coordinates of maximum deflection blade when the flutter phenomenon happened.

Appendix 2 shows the deflected blade coordinates given by MCD.

#### **5. AERODYNAMIC ANALYSIS TO THE SHOCK INSTABILITY PROBLEM**

In order to compare the deflected blade with the original blade, the blade profiles and channels need to be plotted to both blades.

##### **5.1. DEFLECTED BLADE**

For convenience, only three sections were chosen for hub, mean and tip sections from the deflected blade coordinates.

Due to the cylindrical sections, the average radius was taken for the three sections.

The spaces from blade to blade were calculated by the formula.

$$S_i = 2\pi R_i / \text{BLADE COUNT}, \text{ where blade count is } 50.$$

The detailed procedure of calculation and programs are in appendix 3.

## **5.2. ORIGINAL DESIGN BLADE**

Taking the three average radius of hub, mean and tip as interpolation sections in LABLAD, the blades were plotted. The input of LABLAD and plot are in appendix 4.

## **5.3. SHOCK INSTABILITY**

Overlapping the two plots in appendix 3&4, it is easy to see that the downstream parts of blade are deflected down when the shock wave is there.

The shock wave occurs when the critical Mach no. is greatly larger than one ( $M^*$  is 1.498 here). Due to the shock wave, the blades deform to widen the throat area and  $M_1^*$  will decrease (to  $M_2^*$ ) when the shock isn't there.

### **5.3.1. $M_2^*$ CLOSE TO $M_1^*$**

By the blade plots in appendix 3&4, we can easily measure the throat areas and calculate the percent throat area increases as table 1 shown. From Mach table and the calculated throat area ( $A/A^*$ ) changes,  $M_2^*$  can be determined.

TABLE 1: The Percent Throat Area Increase When The Blade Deforms

RADIUS	$M_1^*$	THROAT WIDTH			$M_2^*$
		ORIGINAL	DEFLECTED	% INCREASED	
6.4609		3.75	3.71	-1.07*	
9.72132	1.49	4.73	4.79	1.27	1.48
8.32277	1.46	5.49	5.75	4.74	1.42

Mass Aug. 1.498

From table 1,  $M_2^*$  is close to  $M_1^*$ , and the percent throat area increases are under 10.

Generally speaking, if the throat area change is under 10%, it will be no problem for the blade design.

\*The percent throat area increase at hub section should be zero. The negative value here is due to the plot error, in addition, the throat widths were measured directly from the plots.

### 5.3.2. $M_2^*$ CLOSE TO 1

If  $M_2^*$  is close to 1 (i. e. far from  $M_1^*$ ), it is supposed to input the new rotor exit angles of deflected blades in TAPS program for calculating the exact value of  $M_2^*$  and getting the percent throat area increases from Mach table.

Besides, the blade needs to be checked with DENTON to see what the loading looks like when it deforms.



## **6. CONCLUSIONS**

The purpose of this study is to make certain that whether the current blade design has the flutter problems when LPT operates far from the design point. From the analysis results, we know that it is not necessary to modify the current 1042 rotor blades.

However, to this flutter problem, MCD has started to consider the redesigning work for the LPT backup design.

```

1 60000. 1.1750 16 133 312 304 1.2100 393.82 1.0000 807 809 .80000F-02 .33000E-02
BYPASS AIR LT 699.7, OPEN OIL COOLER, ATR VALVE, WRLD= .1286972956494, NOZZLE AREA 5-11-84
ALT 60000.00 MN 1.175 THRUST 134.58 WF 362.368 AFAW 74.42 CUSTOMER INSTALLATION LOSSES
PAM 1.04 V (TAS) 673.94 13FC 246.924 BPR 0.913 AENG 282.68 LP WR/WA 0.000 HP WR/WA 0.000
TAMF -69.70 V (CAS) 239.31 TT4F/C 1644.97 8966.1 N-LP 16040.71.030 AERIT 587.58 LP-HP 6.000 HP-AP 0.000
TAMC -56.50 ETA RAM .9929 TT4.5 1182.77 617.0 N-HP 19851.7 .854 ANIN 393.82 FAN CUST DP.0.000 PRIMARY DP 0.000
*****
COMPONENT COR FLOW P/P EFF EFPDL PCT H COR N AERO HP AERO TQ BETA SURMAR MAP VALUES WF/PS3RT2
FAN TOTAL 87.435 2.267 .6853 .7188 105.07 16370. 456.44 149.46 11.354 58.029 171.181 2.302 .7000 FIG7 = 9.922
HIP(12-17) 41.730 2.745 .7198 .7559 105.07 16370. 610.40 199.86 11.354 32.520 171.191 2.797 .7352 FIG8 = .453
LPC(12-2-3) 20.091 2.895 .8429 .8638 93.09 16747. 797.24 210.92 5.732 19.955 20.102 2.915 .8493 WFMIN= 1.021
HPC(12-4-3) 8.105 1.666 .8204 .8323 96.80 14102. 5.157 23.337 5.157 23.337 8.051 1.667 .8201 FIG10=4.242
HPY(14-1-3) 6.937 3.377 .8768 .8768 101.40 9855. 1329.61 351.77 T0DL4 158.62 20.199 3.309 .9676 S0PFI= .039
LPT(14.5-4.9) 25.003 3.309 .8669 113.76 9126. 1066.84 349.32 T0DL5 505.33 20.199 3.309 1.0000 1.0000
PWT(16.2-6.4) 64.061 1.000 .0000 0.00 1.0000 1.0000 0.00 0.00 T0DL6 1.000 1.000 1.0000 1.0000
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STATION PRESSURE TEMP FLOW ENTHALPY CYCLE POWER LOSSES BURNER (NO MAR) NOZZLE DATA
AMBIENT 1.040 390.0 498.0 93.195 ACC 362.348 PRB 2.239 PRN .713
2 (1) 2.424 498.0 14.717 119.035 319 CDB .9458 CDC .7941
12(T1P) 2.424 498.0 7.024 6.880 1.0000 CFB .9317 CDF1.0169
(19) 5.494 689.1 6.880 1.0000 AFA 1.492
13 (16) 5.494 689.1
2 (1WB) 2.424 498.0
2.1 (2) 0.654 72.0 LEAKAGE
2.2 (3) 0.654 72.0 .0700 FAN .0020
2.3 (4) 19.260 982.5 .0000 LPC .0010
2.4 (5) 19.260 982.5 1.977 HPC .0020
3 (6) 32.097 982.5 1.977 HPT .0000
3.9 31.283 982.5 1.977 HPT .0000
4.1 (7) 30.283 982.5 1.977 HPT .0000
4.3 (8) 8.967 168.0 1.977 HPT .0000
4.4 (9) 0.967 168.0 1.977 HPT .0000
4.5(10) 0.967 168.0 1.977 HPT .0000
5.1(11) 2.710 1223.3 1.977 HPT .0000
5.1(13) 2.563 1229.2 1.977 HPT .0000
5.9(114) 1.899 1229.2 1.977 HPT .0000
6 (18) 3.563 982.5 1.977 HPT .0000
6.3(19) 2.339 982.5 1.977 HPT .0000
8 (20) 2.328 982.5 1.977 HPT .0000
8 STATIC 1.239 786.0 1.977 HPT .0000
12 STATIC 1.239 786.0 1.977 HPT .0000
*****
REYNOLDS NUMBER CORRECTIONS
COMPO OPT REL EFF P/P FLOW FNTC2 816.00 THETA2 .9601 AFTERBURNER
1 1742 .9790 .9845 9790 .9500 T5FC2 2.7478 RT-THETA2 .9708 WF LB/HP 0.000
2 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EFFICIENCY 1.0000
3 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
4 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
5 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
6 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
7 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
8 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
9 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
10 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
11 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
12 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
*****
REL EFF P/P FLOW FNTC2 816.00 THETA2 .9601 AFTERBURNER
1 1742 .9790 .9845 9790 .9500 T5FC2 2.7478 RT-THETA2 .9708 WF LB/HP 0.000
2 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EFFICIENCY 1.0000
3 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
4 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
5 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
6 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
7 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
8 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
9 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
10 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
11 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
12 1742 .9790 .9845 9790 .9500 NLC2 1659.91 DELTA2 EQVBR .2130
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```



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47800. 1.2400 .77000E-02 32.000 1.4920 373.82 1.0000 .80000E-02 .33000E-02
MAXIMUM INLET CORRECTED AIRFLOW EXCEEDED VALUE RESET TO LIMIT
TFE104Z-70 LPT OPERATION MAX. NOZZLE AREA 5-11-84
ALT 47800.00 MH 1.240 THRUST 291.16 WF 671.735 AFAN 74.42 CUSTOMER INSTALLATION LOSSES
PAN 1.87 V (ICAS) 711.22 TSFC 2.3071 BRP .831 AENG 282.68 LP WB/WA 0.000 HP WB/WA 0.000
TAMF -59.70 V (ICAS) 337.52 IT4/C 1646.67/897.0 N-LP 16236.7/1.042 AXIT 597.58 L6-HP 6.000 HP-HP 0.000
TANC -56.50 ETA RAN .9891 IT4.5 1139.27/615.1 N-HP 20165.7/866 ANIN 393.32 FAN CUST DP,0.000 PRIMARY DP,0.000
*****
COMPONENT COR FLOW P/P EFF EFPLD PCT II COR N AERD HP AERD IQ BETA SURNAR MAP VALUES W/P3SRIZ
FAN TOTAL 88.846 105.07 16370. 2139.63 692.11 105.07 16370. 2139.63 692.11
HIP(12-17) 41.619 2.788 73201 .7301 105.07 16370. 900.91 291.42 11.302 57.525 171.177 2.310 .7009 F1G7 = 94.556
HUB(2-2) 47.228 2.788 73222 .7322 105.07 16370. 1238.72 400.69 11.302 32.343 171.177 2.801 .7361 F1G8 = .181
LPC(12-2-3) 20.440 2.910 8492 .8492 93.42 16807. 1621.93 422.41 6.056 21.608 20.287 2.909 .8489 WFHIN= 1.008
HPC(2-3) 8.198 1.667 8175 .8175 97.21 14161. 1027.73 267.67 5.337 24.727 8.147 1.667 .8175 F1G10=44.416
HPI(4-1-4-3) 6.937 3.395 8792 .8792 102.96 14007. 2680.38 698.11 100L4 156.49 . 6.370 3.395 .8857 F1G11 = .014
LPT(4-3-4-9) 22.144 3.312 8782 .8782 115.28 9248. 2139.63 692.11 100L5 577.72 . 20.188 3.312 .9678 SORTI = .992
PW1(6-2-6-4) 64.216 1.000 1.0000 0.00 ***** 1.000 1.0000 *****
*****
STATION PRESSURE TEMP FLOW ENTHALPY NOZZLE DATA *****
AMBIENT 1.870 390.0 93.3 ***** PR8 2.461 PRM .714
2 (1) 4.725 510.2 13.492 12.2 ***** CDC .7953
12(TIP) 4.725 510.2 13.465 13.465 ***** CDFI.0196
(19) 10.871 706.5 13.468 13.468 ***** A/A 1.492
13 (16) 17.098 708.1 13.477 13.477 *****
2 (HUB) 4.725 510.2 15.228 15.228 ***** LEAKAGE
2.1 (2) 13.174 746.6 13.477 13.477 ***** POLING *****
2.2 (3) 13.174 746.6 13.477 13.477 ***** DCT-HZ .0700 FAN .0020
2.3 (4) 38.331 1051.7 15.228 15.228 ***** HC-4.6 .0000 LPC .001C
2.4 (5) 38.331 1051.7 15.228 15.228 ***** HC-SR .1197 HPC .0020
3 63.906 1242.9 13.477 13.477 ***** HC-RUR .0000 HPI .0000
3.9 60.231 2223.7 12.228 12.228 ***** AIRMR .0044 LPT .0000
4.1 (7) 60.231 2106.3 14.477 14.477 ***** LC-LC .0000
4.3 (8) 17.743 1627.4 15.228 15.228 ***** HC-HC .0000
4.4 (9) 17.743 1598.8 15.228 15.228 ***** *****
4.5(10) 17.743 1598.8 15.228 15.228 ***** *****
6.9(11) 2.358 1226.0 15.228 15.228 ***** *****
5.9(113) 3.749 1222.1 15.434 15.434 ***** *****
6 (18) 5.067 998.4 28.893 28.893 ***** *****
6.3(19) 4.622 988.4 26.871 239.501 ***** *****
8 (20) 4.602 988.4 28.894 239.414 ***** *****
8 STATIC 2.449 770.0 28.894 185.365 ***** *****
12 STATIC 2.449 770.0 28.894 185.365 ***** *****
*****
REYNOLDS NUMBER CORRECTIONS *****
COMPD OPT REL EFF P/P *****
TIP 3287 .9948 *****
HUB 3287 .9948 *****
LPC 3589 1.0004 *****
HPC 1.0000 *****
LPT 2755 .9947 *****
HPI 1.0000 *****
PVT 1.0000 *****
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DATE = 14/05/15 *****
TIME = 14.33 *****
MODEL = *****
WF/P12 142.16 IGV ANGL 1.0 IGVCEFF .7477 EFFPOLY .8188 EXSWIRL .7139

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# LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY



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8350.0      16.      133      304      312      809
      .50000      .77000E-02 37.000      1.4020      393.82      1.0000      .00000E-02 .33000E-02
      MAXIMUM INLET CORRECTED AIRFLOW EXCEEDED VALVE RESET TO LIMIT
      IFEU042-70. LPT OPERATION MAX. NOZZLE AREA 5-11-84
      MN      THRUST      424.36      WF      1036.173      AFAN      76.42      CUSTOMER INSTALLATION LOSSES
      V (TAS)      321.10      TSFC      4.3317      BPR      .961      AENG      282.58      LP W3/W4      0.000      HP 4B/W4      0.000
      V (CAS)      285.52      TT4FC      1656.37 902.4 N-LP      16286.7/1.045      AEXIT      587.58      LP-HP      0.000      HP-HP      0.000
      TARC      -1.54      ETA RAH      1.0000      TT4.5      1146.4/ 610.1 N-HP      20314.7 .874      ANIN      393.82      FAN CUST DP 0.000      PRIMARY DP 0.000
      *****
      COMPONENT      COR FLOW P/P      EFF      EPFDL      PCT I      COR N      AERO HP      AERO TQ      BETA      SURMAR      MAP VALUES      WFP/PS3R2
      FAN TOTAL      89.306      .7026      .7391      105.07      16370.      5843.44      1884.38      .11.208      56.632      171.170      2.323      FLOW      EFF      FIG7 = 9.597
      HUB(2-2.1)      47.987      2.808      .7376      105.07      16370.      3403.41      786.34      .11.208      32.026      171.170      2.808      FIG8 = .113
      LPC(2.2-2.3)      20.625      2.926      .8484      93.42      16879.      4508.66      1163.58      6.154      22.365      20.479      2.926      FIG9 = 5.592
      HPC(2.4-3)      8.234      1.672      .8172      97.53      14209.      2858.51      738.25      5.359      24.888      0.193      1.672      FIGH = 1.005
      HP(4.1-4.3)      6.937      3.394      .8806      103.48      16058.      7387.13      1909.84      100L4      156.15      .6369      3.394      FIGH = 1.005
      LP(4.5-4.9)      63.433      3.258      .8819      115.37      9255.      5843.72      1884.48      100L5      56.84      20.187      3.258      FIGH = 1.005
      PWT(6.2-6.4)      63.250      1.000      0.00      0.00      1.0000      0.00      0.00      100L6      56.84      20.187      3.258      FIGH = 1.0000
      *****
      STATION      PRESSURE      TEMP      FLOW      ENTHALPY      CYCLE POWER LOSSES      BURNER (NO MAR)      NOZZLE DATA
      AMBIENT      10.771      488.9      116.062      ACC S78      EFF      1.0000      FUEL FLOW      1838.173      PR8      1.201      PPM      .705
      2 (1)      12.776      513.4      78.038      122.728      1.0000      FWP/PS3R2      10.809      CDC      .9175      CDC      .7908
      12(TIP)      15.776      513.4      36.106      122.728      1.0000      (SURG)      1.005      CF8      .9256      CDF1.0116
      13 (16)      29.674      711.8      36.106      122.728      1.0000      .0134      VEL      782.      A/A      1.492
      16 (17)      20.092      711.8      36.106      122.728      1.0000      CFDA      .6132      CFDA      .6132
      2 (HUB)      12.776      513.4      78.038      122.728      1.0000      *****
      2.1 (2)      35.870      711.8      36.106      122.728      1.0000      *****
      2.2 (3)      39.877      711.8      36.106      122.728      1.0000      *****
      2.3 (4)      104.9      711.8      36.106      122.728      1.0000      *****
      2.4 (5)      104.9      711.8      36.106      122.728      1.0000      *****
      3 (6)      175.4      711.8      36.106      122.728      1.0000      *****
      3.9 (7)      165.4      711.8      36.106      122.728      1.0000      *****
      4.1 (7)      165.4      711.8      36.106      122.728      1.0000      *****
      4.3 (8)      48.711      711.8      36.106      122.728      1.0000      *****
      4.4 (9)      48.711      711.8      36.106      122.728      1.0000      *****
      4.5 (10)      48.711      711.8      36.106      122.728      1.0000      *****
      5 (11)      14.953      711.8      36.106      122.728      1.0000      *****
      5.9 (14)      10.612      711.8      36.106      122.728      1.0000      *****
      6.3 (10)      14.166      711.8      36.106      122.728      1.0000      *****
      8 (STATIC)      6.883      949.8      78.257      229.778      *****
      12 (STATIC)      6.883      949.8      78.257      229.778      *****
      *****
      REYNOLDS NUMBER CORRECTIONS
      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
      TIP      .8813      1.0000      1.0000      .9500      FNTC2      488.12      THETA2      .9898      WF LB/HP      0.000
      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      WFP/PT2      143.87      IGV ANGL      1.0      EFFFDLY      .7499
      *****
      REYNOLDS NUMBER CORRECTIONS
      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
      TIP      .8813      1.0000      1.0000      .9500      FNTC2      488.12      THETA2      .9898      WF LB/HP      0.000
      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      DATE = 94/05/10
      TIME = 14:21:02
      CPU TIME = 21.62
      MODEL =
      WFP/PT2      143.87      IGV ANGL      1.0      EFFFDLY      .7499
      *****
      REYNOLDS NUMBER CORRECTIONS
      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
      TIP      .8813      1.0000      1.0000      .9500      FNTC2      488.12      THETA2      .9898      WF LB/HP      0.000
      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      *****
      REYNOLDS NUMBER CORRECTIONS
      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
      TIP      .8813      1.0000      1.0000      .9500      FNTC2      488.12      THETA2      .9898      WF LB/HP      0.000
      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      *****
      REYNOLDS NUMBER CORRECTIONS
      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
      TIP      .8813      1.0000      1.0000      .9500      FNTC2      488.12      THETA2      .9898      WF LB/HP      0.000
      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      CPU TIME = 21.62
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      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      CPU TIME = 21.62
      MODEL =
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      *****
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      PWT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      REYNOLDS NUMBER CORRECTIONS
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      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
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      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      COMPO      OPT      REL      EFF      P/P      FLOW      P-FACTORS
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
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      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
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      HUB      .8813      1.0000      1.0000      1.0000      MFC2      16369.91      RT-THETA2      .8694      EFFICIENCY      1.0000
      LPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      HPC      .8813      1.0000      1.0000      1.0000      MFC2      20314.72      DELTA2      .8694      EFFICIENCY      1.0000
      LPT      .8813      1.0000     
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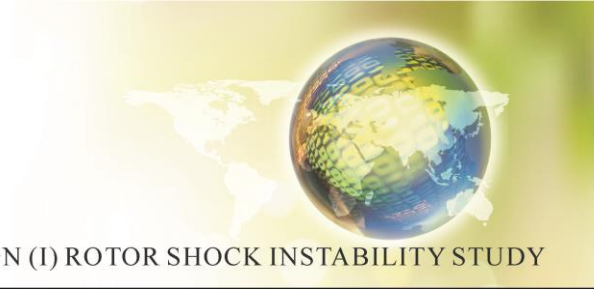




LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY

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3	-.6237319	.6242249E-01	6.466708	180
4	-.5121656	.1444113	6.465391	190
5	-.3909750	.2075828	6.463691	200
6	-.2623143	.2465677	6.462268	210
7	-.1283376	.2559968	6.461782	220
9	.8801007E-02	.2305010	6.462891	230
10	.1050528	.1907664	6.464187	240
11	.1950568	.1382652	6.465520	250
13	.2778141	.7506096E-01	6.466564	260
14	.3489820	.6852357E-02	6.467109	270
15	.4136955	-.6652380E-01	6.466738	280
16	.4727568	-.1442885	6.465390	290
17	.5269681	-.2256628	6.463001	300
18	.5771314	-.3098677	6.459511	310
19	.6240489	-.3961242	6.454857	320
20	.6685229	-.4836533	6.448977	330
21	.7113554	-.5716761	6.441810	340
22	.7533486	-.6594134	6.433293	350
23	-.7790527	-.1726974	6.464694	370
24	-.6917511	-.1256090	6.465755	380
25	-.6005043	-.8344277E-01	6.466443	390
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27	-.4093513	-.2099430E-01	6.466998	410
28	-.3110334	-.4270620E-02	6.467015	420
29	-.2119467	.4136059E-03	6.466956	430
30	-.1128854	-.8720924E-02	6.466898	440
32	-.1464342E-01	-.3345352E-01	6.466913	450
33	.8729288E-01	-.7201336E-01	6.466599	460
34	.1841546	-.1220849	6.465847	470
36	.2752933	-.1818747	6.464442	480
37	.3416686	-.2335919	6.462809	490
38	.4046541	-.2890335	6.460554	500
39	.4645383	-.3477317	6.457637	510
40	.5216099	-.4092187	6.454022	520
41	.5761573	-.4730265	6.449670	530
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44	.7275404	-.6736977	6.431813	560
169	-.7756061	.3659302	8.324962	1800
170	-.6865524	.4350246	8.321206	1810
171	-.5903648	.4739113	8.319255	1820
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173	-.3869397	.4756692	8.319541	1840
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175	-.1860352	.4004201	8.323374	1860
176	-.9299822E-01	.3430480	8.325857	1870
177	-.8355518E-02	.2773801	8.328310	1880
179	.6530488E-01	.2070686	8.330427	1890
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181	.2382893	.3095247E-02	8.333052	1910
182	.3166412	-.1050435	8.332338	1920
183	.3900859	-.2166183	8.330145	1930
184	.4590130	-.3311299	8.326379	1940
185	.5238122	-.4480793	8.320944	1950
186	.5848732	-.5669675	8.313746	1960
187	.6425856	-.6872953	8.304689	1970
188	.6973391	-.8085638	8.293679	1980
189	-.7526228	.3312706	8.326413	2000

190	-.6570443	.3682292	8.324722	2010
191	-.5624022	.3798637	8.324274	2020
192	-.4694494	.3697526	8.324793	2030
193	-.3789386	.3414748	8.326004	2040
194	-.2916227	.2986090	8.327633	2050
195	-.2082545	.2447339	8.329405	2060
196	-.1295870	.1834282	8.331046	2070
197	-.5637298E-01	.1182708	8.332280	2080
199	.1063476E-01	.5284037E-01	8.332832	2090
200	.9383093E-01	-.3678915E-01	8.332970	2100
201	.1737535	-.1291419	8.332035	2110
202	.2507808	-.2238540	8.329993	2120
203	.3252913	-.3205617	8.326806	2130
204	.3976635	-.4189010	8.322438	2140
205	.4682758	-.5185081	8.316853	2150
206	.5375066	-.6190191	8.310014	2160
207	.6057345	-.7200702	8.301884	2170
208	.6733377	-.8212976	8.292428	2180
289	-.5731527	.6342002	9.712316	3000
290	-.4975688	.6402898	9.711916	3010
291	-.4221559	.6308890	9.712533	3020
293	-.3505483	.6060423	9.714114	3030
294	-.2656971	.5547343	9.717227	3040
295	-.1909660	.4915809	9.720578	3050
296	-.1238885	.4197815	9.723910	3060
297	-.6199852E-01	.3425359	9.726969	3070
298	-.2829597E-02	.2630435	9.729501	3080
300	.5608455E-01	.1845040	9.731251	3090
301	.1189454	.8910823E-01	9.732598	3100
302	.1800427	-.7424830E-02	9.733002	3110
303	.2394024	-.1050324	9.732433	3120
304	.2970501	-.2036517	9.730866	3130
305	.3530116	-.3032200	9.728272	3140
306	.4073125	-.4036744	9.724625	3150
307	.4599786	-.5049522	9.719897	3160
308	.5110357	-.6069906	9.714061	3170
309	.5605095	-.7097269	9.707089	3180
310	-.5701108	.6076255	9.714015	3200
311	-.4925392	.5944324	9.714831	3210
312	-.4181775	.5684612	9.716388	3220
314	-.3489876	.5310331	9.718503	3230
315	-.2747088	.4756884	9.721400	3240
316	-.2076407	.4127442	9.724244	3250
317	-.1459920	.3442277	9.726891	3260
318	-.8797086E-01	.2721656	9.729194	3270
319	-.3178583E-01	.1985851	9.731009	3280
321	.2435471E-01	.1255130	9.732191	3290
322	.8657294E-01	.3460166E-01	9.732967	3300
323	.1466899	-.5763706E-01	9.732849	3310
324	.2050492	-.1509742	9.731829	3320
325	.2619946	-.2451808	9.729897	3330
326	.3178696	-.3400279	9.727044	3340
327	.3730178	-.4352865	9.723262	3350
328	.4277830	-.5307276	9.718539	3360
329	.4825087	-.6261223	9.712869	3370
330	.5375385	-.7212416	9.706240	3380



LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY

---

```
.PROC,DBLADES.  
GET,TAPE4=DTAPE4,DTAPE7.  
FTN,I=DTAPE7,L=0,OPT=0.  
LGO.  
GET,PLOTB1,DPLOTI.  
REWIND,*.  
PLOTREQ,DEST=Y,PEN=3,PA=TG.  
PLOTB1,DPLOTI,P.  
REVERT.---PLOTIT,PLOT.---
```

```
PROGRAM DRAVG(INPUT,OUTPUT,TAPE4,TAPE5,TAPE7)
  DIMENSION RH(40),RM(40),RT(40),XH(40),XM(40),XT(40),
1  YH(40),YM(40),YT(40),SH(40),SM(40),ST(40),
2  YH2(40),YM2(40),YT2(40)
  PI=3.1415926
  READ(4,10)(XH(I),YH(I),RH(I),I=1,40)
10  FORMAT(6X,3E15.7)
  READ(4,10)(XM(I),YM(I),RM(I),I=1,38)
  READ(4,10)(XT(I),YT(I),RT(I),I=1,38)
  CALL AVG(RH,40,VMH)
  CALL AVG(RM,38,VMM)
  CALL AVG(RT,38,VMT)
  WRITE(5,40)VMH,VMM,VMT
40  FORMAT(3F10.5)
  DO 50 I=1,20
  SH(I)=2.*PI*RH(I+20)/50.
  SM(I)=2.*PI*RM(I+18)/50.
  ST(I)=2.*PI*RT(I+18)/50.
50  CONTINUE
  DO 60 I=1,20
  YH2(I+20)=YH(I+20)+SH(I)
  YM2(I+18)=YM(I+18)+SM(I)
  YT2(I+18)=YT(I+18)+ST(I)
60  CONTINUE
  WRITE(7,10)(XH(I),YH(I),RH(I),I=1,40)
  WRITE(7,10)(XH(I),YH2(I),RH(I),I=21,40)
  WRITE(7,10)(XM(I),YM(I),RM(I),I=1,38)
  WRITE(7,10)(XM(I),YM2(I),RM(I),I=20,38)
  WRITE(7,10)(XT(I),YT(I),RT(I),I=1,38)
  WRITE(7,10)(XT(I),YT2(I),RT(I),I=20,38)
  END
  SUBROUTINE AVG(A,N,VM)
  DIMENSION A(40)
  T=0.
  DO 1 I=1,N
1  T=T+A(I)
  VM=T/FLOAT(N)
  RETURN
  END
```

DTAPE7



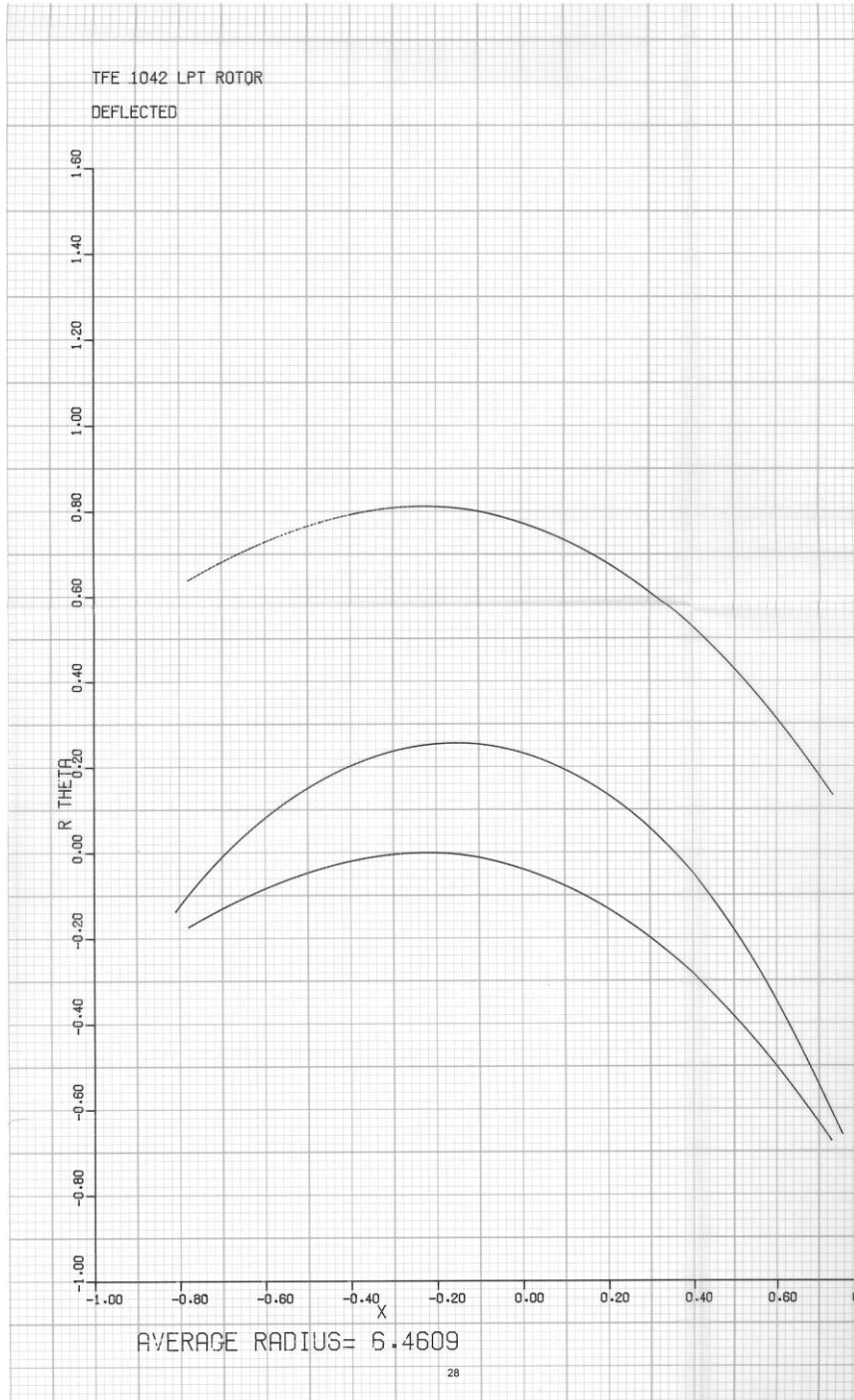
LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY

```
TURBINE 9-4-84 PAGE 3 1 0
TFE 1042 LPT ROTOR
DEFLECTED
AVERAGE RADIUS= 6.4609
3,20,9.,13.,0,1,1,0
X R THETA
```

```
000
(6X,2E15.7)
(20(/),(6X,2E15.7))
(40(/),(6X,2E15.7))
-1.0,0.2,-1.0,0.2
TFE 1042 LPT ROTOR
DEFLECTED
AVERAGE RADIUS= 8.32277
3,19,9.,13.,0,1,1,0
X R THETA
```

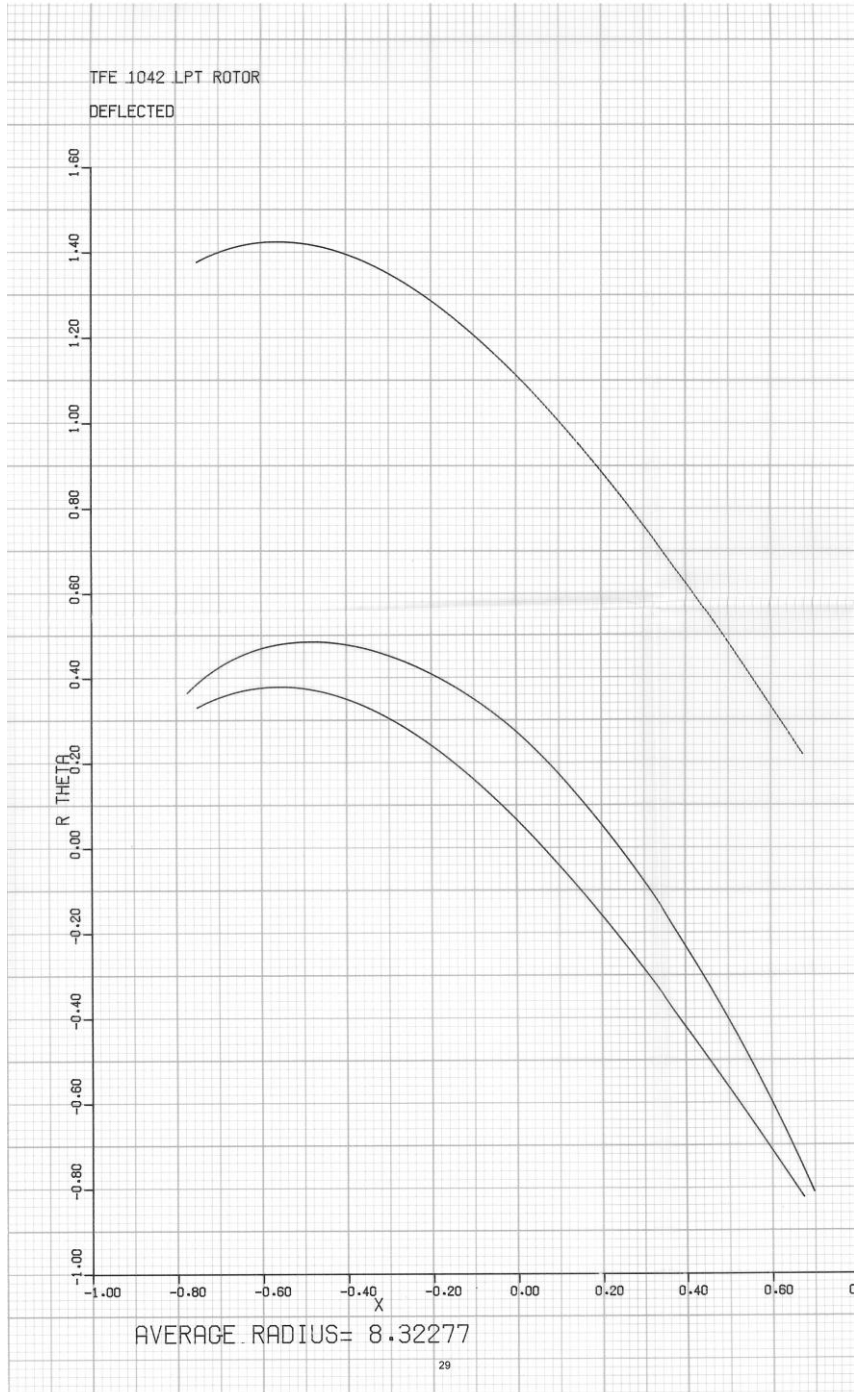
```
000
(60(/),(6X,2E15.7))
(79(/),(6X,2E15.7))
(98(/),(6X,2E15.7))
-1.0,0.2,-1.00,0.2
TFE 1042 LPT ROTOR
DEFLECTED
AVERAGE RADIUS= 9.72132
3,19,9.,13.,0,1,1,0
X R THETA
```

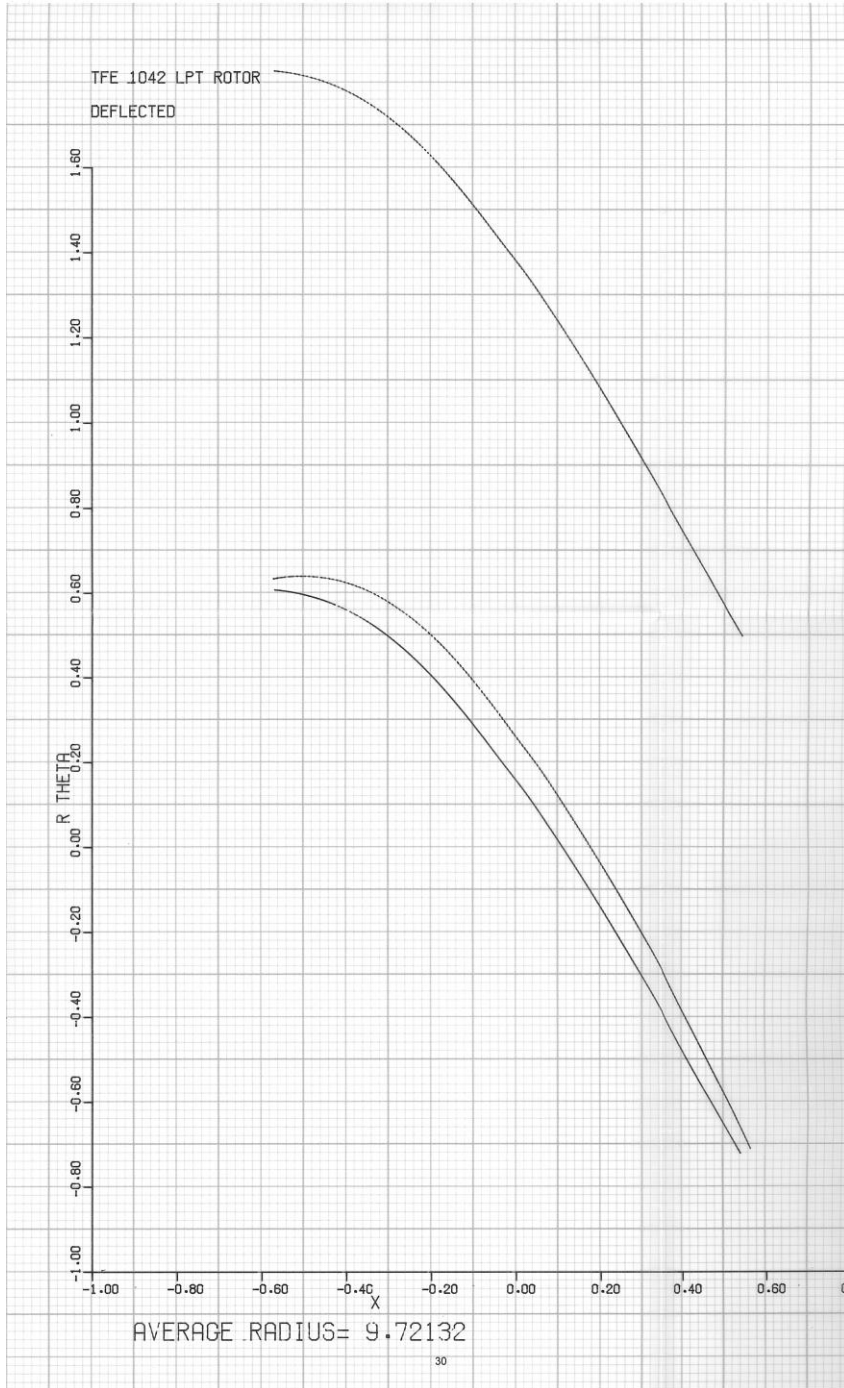
```
000
(117(/),(6X,2E15.7))
(136(/),(6X,2E15.7))
(155(/),(6X,2E15.7))
-1.0,0.2,-1.00,0.2
```





# LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY







9.46886	1.00000	4.80385						780
9.49053	.97066	.23761						790
.43803	.81336	.67949	.99155	0.000001				800
TFE1042-70	LOW PRESSURE	TURBINE			ROTOR	1 AT RIN =		810
1.32133	50.00000	.02500	62.34264	2.00000	8.50000	1.00843		820
.49872								830
10.03722	1.03886	5.02416						840
10.08100	1.00000	3.01452						850
10.08100	.95142	0.00000						860
.45585	.80962	.72005	.98170	0.000001				870



# LPT BACKUP DESIGN (I) ROTOR SHOCK INSTABILITY STUDY

